



Designation: D6873/D6873M – 19

# Standard Practice for Bearing Fatigue Response of Polymer Matrix Composite Laminates<sup>1</sup>

This standard is issued under the fixed designation D6873/D6873M; the number immediately following the designation indicates the year of original adoption or, in the case of revision, the year of last revision. A number in parentheses indicates the year of last reappraisal. A superscript epsilon ( $\epsilon$ ) indicates an editorial change since the last revision or reappraisal.

## 1. Scope

1.1 This practice provides instructions for modifying static bearing test methods to determine the fatigue behavior of composite materials subjected to cyclic bearing forces. The composite material forms are limited to continuous-fiber reinforced polymer matrix composites in which the laminate is both symmetric and balanced with respect to the test direction. The range of acceptable test laminates and thicknesses are described in 8.2.

1.2 This practice supplements Test Method D5961/D5961M with provisions for testing specimens under cyclic loading. Several important test specimen parameters (for example, fastener selection, fastener installation method, and fatigue force/stress ratio) are not mandated by this practice; however, repeatable results require that these parameters be specified and reported.

1.3 This practice is limited to test specimens subjected to constant amplitude uniaxial loading, where the machine is controlled so that the test specimen is subjected to repetitive constant amplitude force (stress) cycles. Either engineering stress or applied force may be used as a constant amplitude fatigue variable. The repetitive loadings may be tensile, compressive, or reversed, depending upon the test specimen and procedure utilized.

1.4 The values stated in either SI units or inch-pound units are to be regarded separately as standard. The values stated in each system are not necessarily exact equivalents; therefore, to ensure conformance with the standard, each system shall be used independently of the other, and values from the two systems shall not be combined.

1.4.1 Within the text the inch-pound units are shown in brackets.

1.5 *This standard does not purport to address all of the safety concerns, if any, associated with its use. It is the responsibility of the user of this standard to establish appropriate*

<sup>1</sup> This practice is under the jurisdiction of ASTM Committee D30 on Composite Materials and is the direct responsibility of Subcommittee D30.05 on Structural Test Methods.

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*appropriate safety, health, and environmental practices and determine the applicability of regulatory limitations prior to use.*

1.6 *This international standard was developed in accordance with internationally recognized principles on standardization established in the Decision on Principles for the Development of International Standards, Guides and Recommendations issued by the World Trade Organization Technical Barriers to Trade (TBT) Committee.*

## 2. Referenced Documents

2.1 *ASTM Standards:*<sup>2</sup>

D883 Terminology Relating to Plastics

D3878 Terminology for Composite Materials

D5229/D5229M Test Method for Moisture Absorption Properties and Equilibrium Conditioning of Polymer Matrix Composite Materials

D5961/D5961M Test Method for Bearing Response of Polymer Matrix Composite Laminates

E4 Practices for Force Verification of Testing Machines

E6 Terminology Relating to Methods of Mechanical Testing

E122 Practice for Calculating Sample Size to Estimate, With Specified Precision, the Average for a Characteristic of a Lot or Process

E177 Practice for Use of the Terms Precision and Bias in ASTM Test Methods

E456 Terminology Relating to Quality and Statistics

E467 Practice for Verification of Constant Amplitude Dynamic Forces in an Axial Fatigue Testing System

E739 Practice for Statistical Analysis of Linear or Linearized Stress-Life ( $S-N$ ) and Strain-Life ( $\epsilon-N$ ) Fatigue Data

E1823 Terminology Relating to Fatigue and Fracture Testing

## 3. Terminology

3.1 *Definitions*—Terminology D3878 defines terms relating to high-modulus fibers and their composites. Terminology D883 defines terms relating to plastics. Terminology E6 defines terms relating to mechanical testing. Terminology E1823 defines terms relating to fatigue. Terminology E456 and

<sup>2</sup> For referenced ASTM standards, visit the ASTM website, [www.astm.org](http://www.astm.org), or contact ASTM Customer Service at [service@astm.org](mailto:service@astm.org). For *Annual Book of ASTM Standards* volume information, refer to the standard's Document Summary page on the ASTM website.

Practice **E177** define terms relating to statistics. In the event of a conflict between terms, Terminology **D3878** shall have precedence over the other standards.

**NOTE 1**—If the term represents a physical quantity, its analytical dimensions are stated immediately following the term (or letter symbol) in fundamental dimension form, using the following ASTM standard symbology for fundamental dimensions, shown within square brackets:  $[M]$  for mass,  $[L]$  for length,  $[T]$  for time,  $[\theta]$  for thermodynamic temperature, and  $[nd]$  for non-dimensional quantities. Use of these symbols is restricted to analytical dimensions when used with square brackets, as the symbols may have other definitions when used without the brackets.

### 3.2 Definitions of Terms Specific to This Standard:

3.2.1 *bearing force,  $P$   $[MLT^{-2}]$ ,  $n$* —the total force carried by a bearing coupon.

3.2.2 *constant amplitude loading,  $n$ —in fatigue*, a loading in which all of the peak values of force (stress) are equal and all of the valley values of force (stress) are equal.

3.2.3 *fatigue loading transition,  $n$ —in the beginning of fatigue loading*, the number of cycles before the force (stress) reaches the desired peak and valley values.

3.2.4 *force (stress) ratio,  $R$   $[nd]$ ,  $n$ —in fatigue loading*, the ratio of the minimum applied force (stress) to the maximum applied force (stress).

3.2.5 *frequency,  $f$   $[T^{-1}]$ ,  $n$ —in fatigue loading*, the number of force (stress) cycles completed in 1 s (Hz).

3.2.6 *hole elongation,  $\Delta D$   $[L]$ ,  $n$* —the permanent change in hole diameter in a bearing coupon caused by damage formation, equal to the difference between the hole diameter in the direction of the bearing force after a prescribed loading and the hole diameter prior to loading.

3.2.7 *nominal value,  $n$* —a value, existing in name only, assigned to a measurable property for the purpose of convenient designation. Tolerances may be applied to a nominal value to define an acceptable range for the property.

3.2.8 *peak,  $n$ —in fatigue loading*, the occurrence where the first derivative of the force (stress) versus time changes from positive to negative sign; the point of maximum force (stress) in constant amplitude loading.

3.2.9 *residual strength,  $[MLT^{-2}]$ ,  $n$* —the value of force (stress) required to cause failure of a specimen under quasi-static loading conditions after the specimen is subjected to fatigue loading.

3.2.10 *run-out,  $n$ —in fatigue*, an upper limit on the number of force cycles to be applied.

3.2.11 *spectrum loading,  $n$ —in fatigue*, a loading in which the peak values of force (stress) are not equal or the valley values of force (stress) are not equal (also known as variable amplitude loading or irregular loading).

3.2.12 *valley,  $n$ —in fatigue loading*, the occurrence where the first derivative of the force (stress) versus time changes from negative to positive sign; the point of minimum force (stress) in constant amplitude loading.

3.2.13 *wave form,  $n$* —the shape of the peak-to-peak variation of the force (stress) as a function of time.

### 3.3 Symbols:

$d$	= fastener or pin diameter
$D$	= specimen hole diameter
$D_i$	= measured hole diameter prior to fatigue loading
$D_N$	= measured hole diameter after $N$ fatigue cycles
$h$	= specimen thickness
$k$	= calculation factor used in bearing equations to distinguish single-fastener tests from double-fastener tests
$K_i$	= joint stiffness prior to fatigue loading
$K_N$	= joint stiffness after $N$ fatigue cycles
$N$	= number of constant amplitude cycles
$P$	= force carried by specimen
$P^{max}$	= greater of the absolute values of the peak and valley values of force
$P^{min}$	= lesser of the absolute values of the peak and valley values of force
$\delta$	= crosshead or extensometer translation
$\delta_i$	= fastener translation prior to fatigue loading
$\delta_N$	= fastener translation after $N$ fatigue cycles
$\delta_{Nc}$	= crosshead or extensometer displacement at zero force after quasi-static compressive loading
$\delta_{Nt}$	= crosshead or extensometer displacement at zero force after quasi-static tensile loading
$\Delta D_N$	= hole elongation after $N$ fatigue cycles
$\Delta K_N$	= percent reduction in joint stiffness after $N$ fatigue cycles
$\Delta P$	= change in force over joint stiffness range under quasi-static loading
$\Delta \delta$	= change in crosshead or extensometer displacement over joint stiffness range under quasi-static loading
$\sigma^{alt}$	= alternating bearing stress during fatigue loading
$\sigma^{brm}$	= maximum cyclic bearing stress magnitude, given by the greater of the absolute values of $\sigma^{max}$ and $\sigma^{min}$
$\sigma^{max}$	= value of stress corresponding to the peak value of force (stress) under constant amplitude loading
$\sigma^{maxq}$	= value of stress corresponding to the peak value of force (stress) under quasi-static loading for measurement of hole elongation and joint stiffness, given by the greater of the absolute values of $\sigma^{max}$ and $0.5 \times \sigma^{min}$
$\sigma^{mean}$	= mean bearing stress during fatigue loading
$\sigma^{min}$	= value of stress corresponding to the valley value of force (stress) under constant amplitude loading
$\sigma^{minq}$	= value of stress corresponding to the valley value of force (stress) under quasi-static loading for measurement of hole elongation and joint stiffness, given by the greater of the absolute values of $\sigma^{min}$ and $0.5 \times \sigma^{max}$

## 4. Summary of Practice

4.1 In accordance with Test Method **D5961/D5961M**, but under constant amplitude fatigue loading, perform a uniaxial test of a bearing specimen. Cycle the specimen between minimum and maximum axial forces (stresses) at a specified frequency. At selected cyclic intervals, determine the hole elongation either through direct measurement or from a force (stress) versus deformation curve obtained by quasi-statically loading the specimen through one tension-compression cycle. If hole elongation is determined from a force (stress) versus

deformation curve, also determine the percent joint stiffness reduction using the force versus deformation data. Determine the number of force cycles at which failure occurs, or at which a predetermined hole elongation or percent joint stiffness reduction is achieved, for a specimen subjected to a specific force (stress) ratio and bearing stress magnitude.

## 5. Significance and Use

5.1 This practice provides supplemental instructions for using Test Method **D5961/D5961M** to obtain bearing fatigue data for material specifications, research and development, material design allowables, and quality assurance. The primary property that results is the fatigue life of the test specimen under a specific loading and environmental condition. Replicate tests may be used to obtain a distribution of fatigue life for specific material types, laminate stacking sequences, environments, and loading conditions. Guidance in statistical analysis of fatigue data, such as determination of linearized stress life (S-N) curves, can be found in Practice **E739**.

5.2 This practice can be utilized in the study of fatigue damage in a polymer matrix composite bearing specimen. The loss in strength associated with fatigue damage may be determined by discontinuing cyclic loading to obtain the static strength using Test Method **D5961/D5961M**.

NOTE 2—This practice may be used as a guide to conduct spectrum loading. This information can be useful in the understanding of fatigue behavior of composite structures under spectrum loading conditions, but is not covered in this standard.

5.3 Factors that influence bearing fatigue response and shall therefore be reported include the following: material, methods of material fabrication, accuracy of lay-up, laminate stacking sequence and overall thickness, specimen geometry, specimen preparation (especially of the hole), fastener-hole clearance, fastener type, fastener geometry, fastener installation method, fastener torque (if appropriate), countersink depth (if appropriate), specimen conditioning, environment of testing, time at temperature, type of mating material, number of fasteners, type of support fixture, specimen alignment and gripping, test frequency, force (stress) ratio, bearing stress magnitude, void content, and volume percent reinforcement. Properties that result include the following:

5.3.1 Hole elongation versus fatigue life curves for selected bearing stress values.

5.3.2 Percent joint stiffness reduction versus fatigue life curves for selected bearing stress values.

5.3.3 Bearing stress versus hole elongation curves at selected cyclic intervals.

5.3.4 Bearing stress versus percent joint stiffness reduction curves at selected cyclic intervals.

5.3.5 Bearing stress versus fatigue life curves for selected hole elongation values.

5.3.6 Bearing stress versus fatigue life curves for selected percent joint stiffness reduction values.

## 6. Interferences

6.1 *Force (Stress) Ratio*—Results are affected by the force (stress) ratio under which the tests are conducted. Specimens loaded under tension-tension or compression-compression force (stress) ratios develop hole elongation damage on one

side of the fastener hole, whereas specimens loaded under tension-compression force (stress) ratios can develop damage on both sides of the fastener hole. Experience has demonstrated that reversed (tension-compression) force ratios are critical for bearing fatigue-induced hole elongation, with fully reversed tension-compression ( $R = -1$ ) being the most critical force ratio **(1-3)**.<sup>3</sup>

6.2 *Loading Frequency*—Results are affected by the loading frequency at which the test is conducted. High cyclic rates may induce heating due to friction within the joint, and may cause variations in specimen temperature and properties of the composite. Varying the cyclic frequency during the test is generally not recommended, as the response may be sensitive to the frequency utilized and the resultant thermal history.

6.3 *Fastener Torque/Pre-load*—Results are affected by the installed fastener pre-load (clamping pressure). Laminates can exhibit significant differences in hole elongation behavior and failure mode due to changes in fastener pre-load under both tensile and compressive loading. Experience has demonstrated that low fastener torque/clamp-up is generally critical for bearing fatigue-induced hole elongation **(1, 2, 4)**. It should be noted that in some instances, low torque testing of single shear specimens has proven unsuccessful due to loosening of the fastener nut/collar during fatigue loading caused by deformation of the pin/bolt.

6.4 *Debris Buildup and Removal*—Results are affected by the buildup of fiber-matrix debris resulting from damage associated with hole elongation, and whether such debris is removed during the test. The presence of debris may mask the actual degree of hole elongation, and can increase both the friction force transfer and temperature within the specimen under fatigue loading. Experience has demonstrated that non-reversed force ratios (especially compression-compression force ratios) exhibit greater debris buildup than reversed force ratios. Fastener and debris removal can facilitate a more accurate measurement of hole elongation **(1, 2, 4)**. In general, removing fastener(s) and cleaning the specimen hole(s) prior to measurement is recommended to ensure conservatism of hole elongation data to account for the potential removal of debris over time (due to fluid exposure, for example). However, fastener and debris removal during the test may result in an unrepresentative measurement of hole elongation growth behavior; thus, fastener and debris removal requirements shall be specified by the test requestor. Fasteners such as blind bolts and lockbolts are not practical to remove during fatigue testing; use of such fasteners may preclude cleaning of the specimen hole(s).

6.5 *Environment*—Results are affected by the environmental conditions under which the tests are conducted. Laminates tested in various environments can exhibit significant differences in hole elongation behavior, joint stiffness response, and failure mode. Experience has demonstrated that elevated temperature, humid environments are generally critical for

<sup>3</sup> The boldface numbers in parentheses refer to the list of references at the end of this standard.

bearing fatigue-induced hole elongation (1-4). However, critical environments must be assessed independently for each material system, stacking sequence, and torque condition tested.

6.6 *Fastener-Hole Clearance*—Bearing fatigue test results are affected by the clearance arising from the difference between hole and fastener diameters. Small changes in clearance can change the number of cycles at which hole elongation initiates, and can affect damage propagation behavior (1). For this reason, both the hole and fastener diameters must be accurately measured and recorded. A typical aerospace tolerance on fastener-hole clearance is  $+75/-0\ \mu\text{m}$  [ $+0.003/-0.000\ \text{in.}$ ] for structural fastener holes.

6.7 *Fastener Type/Hole Preparation*—Results are affected by the geometry and type of fastener utilized (for example, lockbolt, blind bolt) and the fastener installation procedures. Results are also affected by the hole preparation procedures.

6.8 *Method of Hole Elongation and Joint Stiffness Measurement*—Results are affected by the method used to monitor hole elongation and joint stiffness. Direct measurement of hole elongation permits an accurate examination of the extent of damage and elongation local to the hole surface. However, the measured elongation may not be uniform through the thickness of the laminate and may be uneven along the surface of the hole. Additionally, fasteners such as blind bolts and lockbolts are not practical to remove during fatigue testing; use of such fasteners precludes direct measurement of hole elongation. Force versus deformation data provide an “average” through-thickness measurement of hole elongation, as well as an indication of joint stiffness degradation due to damage formation. However, the accuracy of such measurements for hole elongation is affected by factors such as strain indicator accuracy, signal noise and slippage, grip slippage (for crosshead deflection data), friction within the joint specimen, fastener deformation, bearing deformation of load plates, and so forth. In some circumstances, it may be more useful and appropriate to monitor joint stiffness rather than hole elongation, for example when fatigue damage to both the composite laminate(s) and fastener(s) may be expected, or when testing joints with fasteners with high clamp-up forces (for example, lockbolts) which tend to exhibit low levels of hole elongation. It is recommended that joint stiffness changes be monitored using an extensometer unless it is demonstrated that changes measured using crosshead deflection data are consistent with those obtained from extensometer data. This is due to the additional sources of stiffness measurement error inherent to crosshead deflection data (grip slippage, support fixture deformation, and friction).

6.9 *Reuse or Replacement of Fastener(s)*—Results are affected by whether fastener(s) are reused after hole cleaning and elongation measurement, or whether they are replaced. Both techniques have been used in industry, with reuse being the more common practice. Reuse requires less hardware and ensures a constant fastener diameter (and fastener-hole clearance) is maintained. The removal of thread lubricant during repeated torquing can decrease preload for a given torque level; as lower preload produces more hole elongation, reuse should

produce conservative results. Also, fastener degradation is part of the fatigue process, and replacement could be considered non-conservative. However, if the fastener(s) deforms during test, reuse requires that it be replaced in the same “deformed” position as it was prior to removal. Also, replacement ensures that consistent torque and preload levels are used throughout the test. The technique used during fastener re-installation (reuse or replacement) shall be recorded. It is recommended to vary hole inspection intervals to aid in assessing whether fastener removal and reinstallation affects hole elongation behavior.

6.10 *Support Fixture Wear*—Results can be affected by wear and degradation of the holes, fasteners, and pins of the support fixture (when utilized) under repeated use. This is especially important for specimens tested using the Procedure B and Procedure D configurations when compressive loadings are applied, as fixture wear can result in reduced specimen support and stabilization. Ensure the support fixture pins and fasteners are tight tolerance in accordance with Test Method **D5961/D5961M** requirements.

6.11 *Other*—Additional sources of potential data scatter are documented in Test Method **D5961/D5961M**.

## 7. Apparatus

7.1 *General Apparatus*—General apparatus shall be in accordance with Test Method **D5961/D5961M**. The micrometer or gauge used shall be capable of determining the hole and fastener diameters to  $\pm 8\ \mu\text{m}$  [ $\pm 0.0003\ \text{in.}$ ].

7.2 *Testing Machine*—In addition to the requirements described in Test Method **D5961/D5961M**, the testing machine shall be in conformance with Practice **E467** and shall satisfy the following requirements:

7.2.1 *Drive Mechanism and Controller*—The velocity of the movable head shall be capable of being regulated under cyclic force (stress) conditions. The drive mechanism and controller shall be capable of imparting a continuous sinusoidal loading wave form to the specimen. It is important to minimize drift of the fatigue loading away from the maximum and minimum values. Achieving such accuracy is critical in the development of reliable fatigue life data since small errors in loading may result in significant errors in fatigue life. It is recommended that the test controller be equipped with a Test Amplitude controller, capable of monitoring the fatigue forces at least once every three cycles.

7.2.2 *Force Indicator*—The force indicator shall be in compliance with Practices **E4**. The fatigue rating of the force indicator shall exceed the forces at which testing will take place. Additionally, this practice recommends compliance with Practice **E467** for the development of a system dynamic conversion for the verification of specimen forces to within 1 % of true forces.

7.2.3 *Grips*—The grips shall have sufficient fatigue rating for forces at which testing will take place.

7.3 *Support Fixture*—If compressive forces are applied, either during fatigue loading or during quasi-static loading to determine hole elongation, a support fixture shall be used to

stabilize the specimen. The support fixture shall be in accordance with that described in Test Method **D5961/D5961M** Procedure B for single shear specimens, and with that described in Test Method **D5961/D5961M** Procedure D for double shear specimens.

7.4 *Thermocouple and Temperature Recording Devices*, capable of reading specimen temperature to  $\pm 0.5$  °C [ $\pm 1.0$  °F].

## 8. Sampling and Test Specimens

8.1 *Sampling*—For statistically significant data, the procedures outlined in Practice **E122** should be consulted. From the number of tests selected a statistically significant distribution of data should be obtained for a given material, stacking sequence, environment, and loading condition.

8.1.1 *Sample Size for S-N Curve*—The recommended minimum number of specimens in the development of S-N data is described in **Table 1**. A minimum of three different force (stress) levels is recommended in development of S-N data. For additional procedures consult Practice **E739**.

8.2 *Geometry*—In addition to the requirements described in Test Method **D5961/D5961M**, the specimen geometry shall satisfy the following requirements:

8.2.1 *Stacking Sequence*—The stacking sequence should be evaluated for free edge effects to minimize the likelihood of edge delamination initiation.

8.2.2 *Specimen Configuration*—The test specimen configuration shall be in accordance with Test Method **D5961/D5961M** with the following restrictions:

8.2.2.1 *Tensile Loadings Only*—Procedure A (double shear), Procedure B (single shear, two-piece specimen), and Procedure C (single shear, one-piece specimen) configurations may be utilized. For Procedure B, both the single fastener joint and the double fastener joint geometries may be utilized. If the support fixture is used, the length of each specimen half and doubler must be adjusted to accommodate loading with the fixture. Direct measurement of hole diameter(s) is required to determine hole elongation.

8.2.2.2 *Compressive Loadings Applied*—Both the Procedure B (single shear) and Procedure D (double shear) configurations and corresponding support fixtures may be utilized. For Procedure B, both the single fastener joint and the double fastener joint geometries may be utilized; the length of each specimen half and doubler must be adjusted to accommodate loading with the support fixture. Hole elongation may be determined through either direct measurement or quasi-static loadings; joint stiffness may also be determined.

8.2.3 *Adhesive*—For specimens with bonded doublers, the adhesive should have sufficient durability as to withstand fatigue loading for the duration of the test.

**TABLE 1 Number of Specimens Required for Each S-N Curve**

Type of Test	Minimum Number of Test Specimens
Preliminary and exploratory	6
Research and development testing	12
Design allowables data	24
Reliability data	24

8.3 *Specimen Preparation*—Specimens shall be prepared in accordance with Test Method **D5961/D5961M**. Special care should be taken to ensure that specimen edges are sufficiently free of obvious flaws as determined by visual inspection. Such flaws may lead to premature failure due to edge delamination.

## 9. Calibration

9.1 The accuracy of all measuring equipment shall have certified calibrations that are current at the time of use of the equipment.

## 10. Conditioning

10.1 The recommended pre-test condition is effective moisture equilibrium at a specific relative humidity as established by Test Method **D5229/D5229M**; however, if the test requestor does not explicitly specify a pre-test conditioning environment, no conditioning is required and the test specimens may be tested as prepared.

10.2 The pre-test specimen conditioning process, to include specified environmental exposure levels and resulting moisture content, shall be reported with the test data.

NOTE 3—The term moisture, as used in Test Method **D5229/D5229M**, includes not only the vapor of a liquid and its condensate, but the liquid itself in large quantities, as for immersion.

10.3 If no explicit conditioning process is performed, the specimen conditioning process shall be reported as “unconditioned” and the moisture content as “unknown.”

10.4 Maintaining testing environment is critical to obtaining consistent fatigue data since testing for long periods of time (days or weeks) is not uncommon. For unattended tests, the test environment shall be monitored so that unintended changes in the test environment result in suspension of the test. Report the testing environment for the duration of the test.

## 11. Procedure

### 11.1 *Parameters to Be Specified Prior to Test:*

11.1.1 The specimen sampling method, specimen type and geometry, minimum and maximum test forces (stresses)  $\sigma^{min}$  and  $\sigma^{max}$  for each test, force (stress) ratio for each test, test frequency and wave form of the fatigue loading. For the purpose of development of an S-N curve, all specimens shall be tested at the same frequency and wave form unless that is a factor to be studied in the test.

11.1.2 Fatigue cycle counts at which hole elongation (and joint stiffness if applicable) is to be measured, method of measuring hole elongation, fastener and debris removal requirements, quasi-static peak and valley forces for hole elongation and joint stiffness measurement (if applicable), hole elongation level or percent joint stiffness reduction at which fatigue loading shall cease, and run-out cycles. Historically, bearing fatigue testing has ceased after the hole elongation level has reached 10 to 25 % of the initial hole diameter, or after joint stiffness has been reduced by 10 to 25 % from the initial stiffness.

11.1.3 All other parameters documented in Test Method **D5961/D5961M**.

11.2 General Instructions:

11.2.1 Any deviations from these procedures, whether intentional or inadvertent, shall be reported.

11.2.2 Perform general instructions for conditioning, measurement, cleaning, assembly, and fastener torquing in accordance with Test Method D5961/D5961M.

11.3 Test Procedure:

11.3.1 Supported Specimen Installation—If the specimen is to be tested with support fixture, install the test specimen into the support fixture as described in Test Method D5961/D5961M.

11.3.2 Temperature Monitoring—Attach temperature recording device in a manner not to influence the dynamic response of the specimen. It is recommended to attach the device to a fastener, as fatigue loading will typically cause a greater increase in fastener temperature than in laminate temperature. The device may be attached to the fastener using adhesive, tape, or a spring clip; when utilizing a spring clip, use insulating material to isolate the temperature recording device from the spring clip. The temperature of the specimen shall be monitored, and the frequency should be kept low enough to avoid significant temperature variations, unless that is a factor to be studied in the test. Caution is recommended when selecting loading frequencies; high cyclic rates may cause variations in specimen temperature and properties of the composite. For some material systems a change in 10 °C [18 °F] has demonstrated measurable degradation of material properties.

NOTE 4—When testing a conditioned specimen at elevated temperature with no fluid exposure control, the percentage moisture loss of the specimen prior to test completion may be estimated by placing a conditioned traveler coupon of known weight within the test chamber at the same time the specimen is placed in the chamber. Upon completion of the test, the traveler coupon is removed from the chamber, weighed, and the percentage weight loss calculated and reported. It should be noted that specimen moisture loss may differ from the traveler moisture loss due to cyclic loading-induced heating of the specimen.

11.3.3 Specimen Insertion—In accordance with Test Method D5961/D5961M, insert the specimen and support fixture (as applicable) into the test machine.

NOTE 5—Monitor the specimen for the occurrence of slippage or crushing as a result of the grips or fixture. Should either slippage or crushing occur and lead to premature specimen failure, this data should not be reported as valid.

11.3.4 Extensometer Installation—Attach extensometer(s) to the edges of the specimen in accordance with Test Method D5961/D5961M.

NOTE 6—It is recommended that joint stiffness changes be monitored using an extensometer. Crosshead deflection data may be used if it is first demonstrated that percent joint stiffness reduction measurements are consistent with those obtained from extensometer data.

11.3.5 Quasi-Static Loading—If force versus deformation data is being used to determine hole elongation and joint stiffness, perform an initial quasi-static loading cycle.

11.3.5.1 Quasi-Static Forces—The quasi-static tension and compression forces shall be those corresponding to  $\sigma^{maxq}$  and  $\sigma^{minq}$  as defined in 3.3.

11.3.5.2 Loading—From zero force, apply tensile force to the specimen quasi-statically up to the force (stress) corresponding to  $\sigma^{maxq}$ , then return to zero force. Apply compressive force to the specimen up to the force (stress) corresponding to  $\sigma^{minq}$ , then return to zero force. Force (stress) versus crosshead deflection and extensometer deflection shall be recorded during the quasi-static force cycle. A hysteresis curve, similar to those shown in Fig. 1, should be observed after graphically plotting the force (stress) versus deflection data. The quasi-static loading should be conducted under force control with a low loading rate (such that a typical hysteresis cycle takes approximately 20 to 30 s to complete). A minimum sampling rate of 2 to 3 data recordings per second, and a target minimum of 50 data points per hysteresis cycle, are recommended.

NOTE 7—In some instances, the applied tensile and compressive forces may not be high enough to overcome friction at the fastener hole. When this occurs, hole elongation will be relatively small, and it is recommended to measure hole elongation directly (if possible) in addition to taking force versus deformation data.

11.3.5.3 Extensometer Removal—Remove extensometer(s) from the specimen prior to fatigue loading.

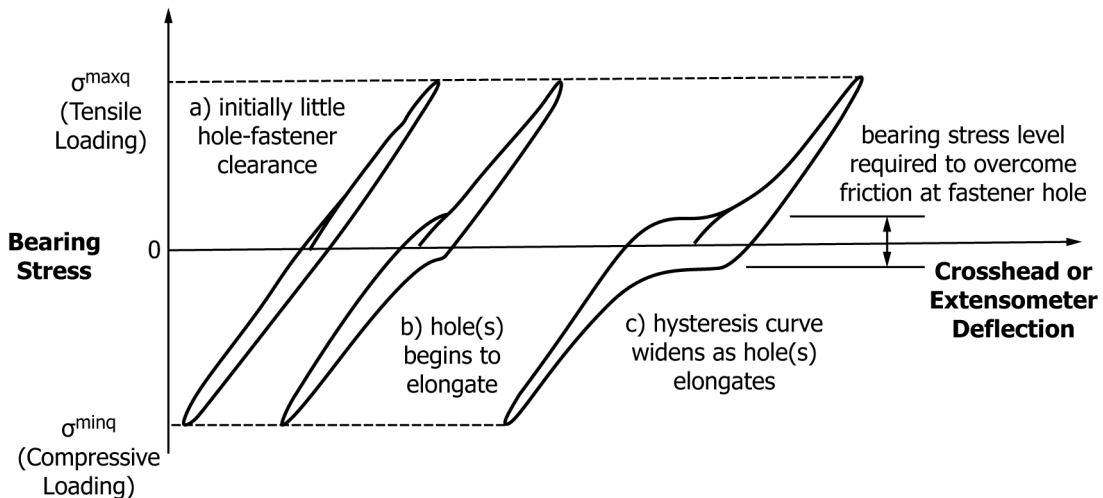


FIG. 1 Typical Bearing Stress versus Deflection Plots Depicting Shape of Hysteresis Curve