

## SLOVENSKI STANDARD SIST EN 6034:2016

01-februar-2016

Aeronavtika - Z ogljikovimi vlakni ojačeni polimerni materiali - Preskusna metoda - Ugotavljanje energije pri medplastni lomni žilavosti - Način II - GIIC

Aerospace series - Carbon fibre reinforced plastics - Test method - Determination of interlaminar fracture toughness energy - Mode II - GIIC

Luft- und Raumfahrt - Kohlenstofffaserverstärkte Kunststoffe - Prüfverfahren - Bestimmung der interlaminaren Energiefreisetzungsrate - Mode II - GIIC

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Série aérospatiale - Matières plastiques renforcées de fibres de carbone - Méthode
d'essai - Détermination de l'énergie de ténacité en rupture interlaminaire - Mode II - GIIC

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Ta slovenski standard je istoveten z: EN 6034-2015

ICS:

49.025.40 Guma in polimerni materiali Rubber and plastics 83.120 Ojačani polimeri Reinforced plastics

SIST EN 6034:2016 en,fr,de

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EUROPEAN STANDARD NORME EUROPÉENNE EUROPÄISCHE NORM **EN 6034** 

November 2015

ICS 49.025.40

#### **English Version**

## Aerospace series - Carbon fibre reinforced plastics - Test method - Determination of interlaminar fracture toughness energy - Mode II - GIIC

Série aérospatiale - Matières plastiques renforcées de fibres de carbone - Méthode d'essai - Détermination de l'énergie de ténacité en rupture interlaminaire - Mode II - GIIC

Luft- und Raumfahrt - Kohlenstofffaserverstärkte Kunststoffe - Prüfverfahren - Bestimmung der interlaminaren Energiefreisetzungsrate - Mode II - GIIC

This European Standard was approved by CEN on 10 August 2013.

CEN members are bound to comply with the CEN/CENELEC Internal Regulations which stipulate the conditions for giving this European Standard the status of a national standard without any alteration. Up-to-date lists and bibliographical references concerning such national standards may be obtained on application to the CEN-CENELEC Management Centre or to any CEN member.

This European Standard exists in three official versions (English, French, German). A version in any other language made by translation under the responsibility of a CEN member into its own language and notified to the CEN-CENELEC Management Centre has the same status as the official versions.

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EUROPEAN COMMITTEE FOR STANDARDIZATION COMITÉ EUROPÉEN DE NORMALISATION EUROPÄISCHES KOMITEE FÜR NORMUNG

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## **European foreword**

This document (EN 6034:2015) has been prepared by the Aerospace and Defence Industries Association of Europe - Standardization (ASD-STAN).

After enquiries and votes carried out in accordance with the rules of this Association, this Standard has received the approval of the National Associations and the Official Services of the member countries of ASD, prior to its presentation to CEN.

This European Standard shall be given the status of a national standard, either by publication of an identical text or by endorsement, at the latest by May 2016, and conflicting national standards shall be withdrawn at the latest by May 2016.

Attention is drawn to the possibility that some of the elements of this document may be the subject of patent rights. CEN [and/or CENELEC] shall not be held responsible for identifying any or all such patent rights.

According to the CEN-CENELEC Internal Regulations, the national standards organizations of the following countries are bound to implement this European Standard: Austria, Belgium, Bulgaria, Croatia, Cyprus, Czech Republic, Denmark, Estonia, Finland, Former Yugoslav Republic of Macedonia, France, Germany, Greece, Hungary, Iceland, Ireland, Italy, Latvia, Lithuania, Luxembourg, Malta, Netherlands, Norway, Poland, Portugal, Romania, Slovakia, Slovenia, Spain, Sweden, Switzerland, Turkey and the United Kingdom.

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## 1 Scope

This standard specifies the procedure to determine the mode II interlaminar fracture toughness energy  $G_{\text{IIC}}$  of carbon fibre composites manufactured from unidirectional tape or woven fabric.

This standard does not give any directions necessary to meet health and safety requirements. It is the responsibility of the user of this standard to consult and establish appropriate health and safety precautions.

#### 2 Normative references

The following documents, in whole or in part, are normatively referenced in this document and are indispensable for its application. For dated references, only the edition cited applies. For undated references, the latest edition of the referenced document (including any amendments) applies.

EN 2565, Aerospace series — Preparation of carbon fibre reinforced resin panels for test purposes 1)

EN 2743, Aerospace series — Fibre reinforced plastics — Standard procedures for conditioning prior to testing unaged materials

EN 2823, Aerospace series — Fibre reinforced plastics — Test method for the determination of the effect of exposure to humid atmosphere on physical and mechanical characteristics <sup>1)</sup>

EN 6033, Aerospace series — Carbon fibre reinforced plastics — Test method — Determination of interlaminar fracture toughness energy (Mode (1-20), 11teh. a1)

#### 3 Terms and definitions

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For the purposes of this document, the following terms and definitions apply.

#### 3.1

## interlaminar fracture toughness energy

the interlaminar fracture toughness energy is the energy per unit plate width which is necessary to produce an unit crack growth at an interlaminar crack between two plies of a laminate

#### 3.2

#### mode II

the mode indicates the method by which the load is applied to produce the crack. Mode Il crack extends as a result of shear forces at the crack tip introduced by a flexural test

#### 3.3

#### GIIC

 $G_{\text{\sc IIC}}$  is the designation of the interlaminar fracture toughness energy determined by a mode II test procedure

<sup>1)</sup> Published as ASD-STAN Prestandard at the date of publication of this standard. <a href="http://www.asd-stan.org/">http://www.asd-stan.org/</a>

## 4 Principle of the method

The end notched flexure specimen is used for mode Il testing. A precracked specimen is loaded in a three point bend fixture (see Figure 2) until crack propagation onset. The load applied to the specimen and the cross head displacement of the test machine are recorded continuously during the test. The total fracture toughness energy is calculated from the initial crack length and load-displacement diagram.

## 5 Designation of the method

The designation of the method used shall be drawn up according to the following example:

Description block

Carbon fibre reinforced plastic materials
Determination of interlaminar fracture toughness energy

Mode II - G<sub>IIC</sub> - test

Number of this standard-

NOTE If necessary, the code I9005 may be placed between the description block and the identity block.

## 6 Apparatus

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**6.1** Test machine accurate to within 1 <u>%in the load range</u> used.

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- **6.2** Device for recording the deflection as a function of load accurate to within 1 % in the deflection range used.
- **6.3** Vernier caliper accurate to the nearest 0,1 mm.
- **6.4** Flat face micrometer accurate to the nearest 0,01 mm.
- **6.5** Microscope of a magnification of 15-25.

## 7 Test specimen

#### 7.1 Test specimen description

To achieve reproducible test results the  $G_{\text{IIC}}$ -test shall only be performed with specimens (see Figure 1) where the initial crack has been introduced in a defined mode I procedure (see EN 6033).

The test specimen shall be cut from the residual part of  $G_{IC}$  specimen tested or at least loaded and cracked per EN 6033, and fulfilling the following requirements.

## 7.2 Test specimen preparation

Prepare the required laminates according to EN 2565 method B for carbon.

The crack front remains approximately a straightline perpendicular to the longitudinal direction of the specimen (as presumed by the marks performed on both already painted longitudinal side faces of the specimen to locate the crack tips after crack stabilization under the mode I loading).

The uncracked portion of the specimen is of sufficient length to comply with the requirements of Figure 1.

The observation during the mode I loading did not show neither fibre bridgings nor moving of the crack to plies adjacent to the mid plies.

The specimen shall be cut in a way that the specified remaining crack length of Figure 1 is guaranteed.

Specimens longer than 112 mm can be used to realize a serie of tests for the determination of  $G_{\text{IIC}}$  if needed.

Appropriate identification shall be maintained throughout the specimen preparation to ensure traceability to the  $G_{IC}$  specimen and to the original laminate.

If not otherwise defined by the specification invoking the test, three specimens shall be tested per test condition. **iTeh STANDARD PREVIEW** 

**CAUTION** — When testing <u>fabric</u> material, the results may be influenced by the position of the crack between the layers (nested, unnested etc.). It shall be clearly stated which interlaminar area shall be tested and how to manufacture the specimens. <u>SIST EN 6034:2016</u>

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## 8 Procedure

#### 8.1 Conditioning

If not otherwise defined by the specification invoking the test, specimens to be tested dry or in the as cured state, should first be conditioned at  $(23 \pm 2)$  °C,  $(50 \pm 5)$  % RH in accordance with EN 2743, same requirements to be applied to perform room temperature testing.

Aged specimens shall be tested directly after ageing procedure (a maximum storage of 8 h at  $(23 \pm 2)$  °C is allowed before testing).

#### 8.2 Determination of dimensions

The width and length of the specimens shall be measured using a vernier caliper (see 6.3).

A minimum of three thickness measurements distributed over the specimen shall be made using a flat face micrometer (see 6.4), the average shall be recorded.

#### 8.3 Testing

The crack length has to be measured by vernier caliper (see 6.3), using the microscope (see 6.5). A thin layer of white ink applied to the longitudinal side faces of the specimen may be used to facilitate this measurement.

For the entire duration of the test, a continuous load-displacement record shall be made, measuring the displacement of the loading nose.

Place the specimen in the test fixture in accordance with Figure 2, assuring the position of the initial crack:

$$a = (35 \pm 1) \,\mathrm{mm}$$

The following procedure shall then apply:

- Adjust the load cell reading at zero.
- Remove any slack between the loading nose and the specimen by making the needed displacement.
- Adjust the displacement reading at zero if necessary.
- Load the specimen under displacement control at a displacement rate of 1 mm/min.
- Optically observe the crack tip in order to detect the crack propagation onset (microscope of 6.5).
- Record the critical load at delamination crack onset and stop the loading as soon as evidence of crack propagation has been confirmed by a small load drop.

## 8.4 Elevated and sub-zero temperature tests

If not otherwise defined by the specification invoking the test, apply a thermocouple on the specimen as close as possible to cracked area of the specimen and seal the thermocouple from the surrounding air by using vacuum bag sealant. (standards.iteh.ai)

The soak time at test temperature, defined by the specification invoking the test, shall be  $(5 \pm 0.5)$  min for dry specimens and in accordance with EN 2823 for conditioned specimen.

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If more then one determination is provided for, wait for crack stabilization under load and mark the current tip location on the painted sides of the specimen.

Unload the specimen to zero.

The complete procedure shall be repeated for the subsequent determinations when needed.

Unless otherwise specified, the remaining uncracked portion of the specimen shall be delaminated after removal from the test fixture (cooling the specimen down to liquid nitrogen temperature will facilitate this operation).

Verification measurements of the crack lengths shall then be made from the crack front (thumbnail marks) on the fractured specimen. Care shall be taken to protect the delamination surfaces for later inspection of failure mode if requested.

The results shall be analysed in accordance with Clause 9.