SLOVENSKI STANDARD
oSIST prEN 16603-10-03:2021

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Vesoljska tehnika - Preskušanje

Space engineering - Testing

Raumfahrttechnik - Tests

Ingénierie spatiale - Vérification par essai

Ta slovenski standard je istoveten z: **prEN 16603-10-03**

ICS:

49.140	Vesoljski sistemi in operacije	Space systems and operations

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English version

Space engineering - Testing

Ingénierie spatiale - Vérification par essai

Raumfahrttechnik - Tests

This draft European Standard is submitted to CEN members for enquiry. It has been drawn up by the Technical Committee CEN/CLC/JTC 5.

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Recipients of this draft are invited to submit, with their comments, notification of any relevant patent rights of which they are aware and to provide supporting documentation.Recipients of this draft are invited to submit, with their comments, notification of any relevant patent rights of which they are aware and to provide supporting documentation.

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European Foreword

This document (prEN 16603-10-03:2021) has been prepared by Technical Committee CEN/CLC/TC 5 “Space”, the secretariat of which is held by DIN (Germany).

This document (prEN 16603-10-03:2021) originates from ECSS-E-ST-10-03C Rev.1 DIR1.

This document is currently submitted to the ENQUIRY.

This document will supersede EN 16603-10-03:2014.

The main changes with respect to EN 16603-10-03:2014 are listed below:

- Implementation of Change Requests received to the ECSS equivalent standard
- xxxx --- The final Change log will be completed before publication --- xxxx

This document has been developed to cover specifically space systems and will therefore have precedence over any EN covering the same scope but with a wider do-main of applicability (e.g.: aerospace).

Introduction

The requirements on the systems engineering process are gathered in ECSS-E-ST-10; while specific aspects are further elaborated in dedicated standards, in particular: ECSS-E-ST-10-06, ECSS-E-ST-10-02 and the present standard (ECSS-E-ST-10-03)

In the System Engineering branch (ECSS-E-10) this standard aims at a consistent application of on ground testing requirements to allow proper qualification and acceptance of space products

Experience has demonstrated that incomplete or improper on ground testing approach significantly increase project risks leading to late discovery of design or workmanship problem(s) or in-orbit failure(s).

Testing is part of the system engineering process as defined in ECSS-E-ST-10. This starts at the early phase of the mission when defining verification process in terms of the model philosophy and sequences of tests and ends at the last testing phase prior launch.

In the level of decomposition of a space system, this standard addresses the requirements for space segment element and space segment equipment.

The document is organised such that:

- clause 4 provides requirements for overall test programme, test management test conditions, test input tolerances and measurement uncertainties;
- clause 5 provides requirements for Space segment equipment;
- clause 6 provides requirements for Space segment element;
- clause 7 provides requirements for Pre-launch testing.

Clauses 5 and 6 are organised as follows:

- general requirements for the products under test applicable to all models (clause 5.1 or 6.1);
- requirements applicable to qualification model (clause 5.2 or 6.2);
- requirements applicable to acceptance model (clause 5.3 or 6.3);
- requirements applicable to protoflight model (clause 5.4 or 6.4);
- detailed implementation requirements (clause 5.5 or 6.5);

In the clause providing requirements for each model (i.e. clauses 5.2, 5.3, 5.4, 6.2, 6.3 and 6.4), the first table of the clause:

- lists all types of test and defines their applicability and conditions;
- links to the second table of the clause that defines tests level and duration;
- provides reference to the clause defining the detailed implementation requirements for the given test (clause 5.5 or 6.5).

For space segment equipment, the required sequence of tests, for each model, is defined by tailoring the two tables in clause 5.2, 5.3 or 5.4.

Since testing activities are part of the overall verification activities, test documentation to be produced (DRD's) are either specified in the ECSS-E-ST-10-02 (case of the test report) or in this document.

Annex D gives guidelines for performing the tailoring of this standard as well as the generation of the compliance and verification matrices.

1 Scope

This standard addresses the requirements for performing verification by testing of space segment elements and space segment equipment on ground prior to launch. The document is applicable for tests performed on qualification models, flight models (tested at acceptance level) and protoflight models.

The standard provides:

- Requirements for test programme and test management,
- Requirements for retesting,
- Requirements for redundancy testing,
- Requirements for environmental tests,
- General requirements for functional and performance tests,

  NOTE Specific requirements for functional and performance tests are not part of this standard since they are defined in the specific project documentation.

- Requirements for qualification, acceptance, and protoflight testing including qualification, acceptance, and proto-fight models' test margins and duration,
- Requirements for test factors, test condition, test input tolerances, and measurement uncertainties,

- General requirements for development tests pertinent to the start of the qualification test programme,

  NOTE Development tests are specific and are addressed in various engineering discipline standards.

- Content of the necessary documentation for testing activities (e.g. DRD).

Due to the specific aspects of the following types of test, this Standard does not address:

- Space system testing (i.e. testing above space segment element), in particular the system validation test,
- In-orbit testing,
- Testing of space segment subsystems,

  NOTE Tests of space segment subsystems are often limited to functional tests that, in some case, are run on dedicated models. If relevant, qualification tests for space segment subsystems are assumed to be covered in the relevant discipline standards.

- Testing of hardware below space segment equipment levels (including assembly, parts, and components),
- Testing of stand-alone software,

NOTE For verification of flight or ground software, ECSS-E-ST-40 and ECSS-Q-ST-80 apply.

- Testing of two-phase heat transport equipment,

  NOTE For acceptance and qualification testing of two-phase heat transport equipment, ECSS-E-ST-31-02 applies.

- Tests of launcher segment, subsystem and equipment, and launch facilities,
- Tests of facilities and ground support equipment,
- Tests of ground segment.

This standard may be tailored for the specific characteristic and constrains of a space project in conformance with ECSS-S-ST-00. Annex D gives guidelines for performing this tailoring.

2
Normative references

The following normative documents contain provisions which, through reference in this text, constitute provisions of this ECSS Standard. For dated references, subsequent amendments to, or revision of any of these publications do not apply. However, parties to agreements based on this ECSS Standard are encouraged to investigate the possibility of applying the more recent editions of the normative documents indicated below. For undated references, the latest edition of the publication referred to applies.

EN reference	Reference in text	Title
EN 16601-00	ECSS-S-ST-00-01	ECSS system - Glossary of terms
EN 16603-10-02	ECSS-E-ST-10-02	Space engineering - Verification
EN 16603-20	ECSS-E-ST-20	Space engineering - Electrical and electronic
EN 16603-20-01	ECSS-E-ST-20-01	Space engineering - Multipactor design and test
EN 16603-20-06	ECSS-E-ST-20-06	Space engineering - Spacecraft charging
EN 16603-20-07	ECSS-E-ST-20-07	Space engineering - Electromagnetic compatibility
EN 16603-20-08	ECSS-E-ST-20-08	Space engineering - Photovoltaic assemblies and components
EN 16603-31	ECSS-E-ST-31	Space engineering - Thermal control general requirements
EN 16603-32	ECSS-E-ST-32	Space engineering - Structural general requirements
EN 16603-32-02	ECSS-E-ST-32-02	Space engineering - Structural design and verification of pressurized hardware
EN 16603-32-10	ECSS-E-ST-32-10	Space engineering - Structural factors of safety for spaceflight hardware
EN 16603-32-11	ECSS-E-ST-32-11	Space engineering - Modal survey assessment
EN 16603-33-01	ECSS-E-ST-33-01	Space engineering - Mechanisms
EN 16601-40	ECSS-M-ST-40	Space project management - Configuration and information management
EN 16602-10-09	ECSS-Q-ST-10-09	Space product assurance - Nonconformance control system
EN 16602-20-07	ECSS-Q-ST-20-07	Space product assurance - Quality assurance for test centres
EN 16602-40	ECSS-Q-ST-40	Space product assurance - Safety
EN 16602-70-01	ECSS-Q-ST-70-01	Space product assurance - Cleanliness and contamination control
	ISO 3740:2000	Acoustics - Determination of sound power levels of noise sources - Guidelines for the use of basic standards

3
Terms, definitions and abbreviated terms

3.1 Terms from other standards

a. For the purpose of this standard; the terms and definitions from ECSS-S-ST-00-01 apply, and in particular the following:

1. flight model
2. lifetime
3. protoflight model
4. qualification model
5. space segment element
6. space segment equipment
7. space segment subsystem
8. structural model
9. system

b. For the purpose of this standard, the following terms and definitions from ECSS-E-ST-10-02 apply:

1. commissioning
2. model philosophy
3. test

c. For the purpose of this Standard, the following terms and definitions from ECSS-E-ST-31 apply:

1. acceptance temperature range
2. design temperature range
3. minimum switch ON temperature
4. predicted temperature range
5. qualification temperature range
6. temperature reference point (TRP)

d. For the purpose of this Standard, the following terms and definitions from ECSS-E-ST-32 apply:

1. burst pressure
2. design burst pressure
3. factor of safety
4. limit load (LL)
5. maximum design pressure (MDP)

6. proof factor
7. proof pressure
8. proof test

3.2 Terms specific to the present standard

3.2.1 24-hour equivalent noise exposure level

equivalent sound pressure level (Leq) to which the crew members are exposed over a 24-hour period; expressed in dBA

NOTE 0 dBA corresponds to 20 μPa.

3.2.2 <<deleted>>

3.2.3 abbreviated functional test (AFT)

See "**reduced functional test (RFT)**"

3.2.4 acceptance level

test level reflecting the maximum level expected to be encountered during the flight product lifetime increased by acceptance margins

3.2.5 acceptance margin

increase of the environmental, mechanical, thermal, electrical, EMC, or operational extremes above the worst case levels predicted over the specified product lifetime for the purpose of workmanship verification

NOTE 1 Margins can include an increase in level or range, an increase in duration or cycles of exposure, as well as any other appropriate increase in severity.

NOTE 2 For thermal acceptance margin refer also to ECSS-E-ST-31.

3.2.6 <<deleted>>

3.2.7 crewed space segment element

space segment design to ensure the safe presence of crew onboard

3.2.8 <<deleted>>

3.2.9 dwell time

duration necessary to ensure that internal parts or subassembly of a space segment equipment have achieved thermal equilibrium, from the start of temperature stabilisation phase, i.e. when the temperature reaches the targeted test temperature plus or minus the test tolerance

3.2.10 environmental tests

tests applied to a product simulating (together or separately) environmental conditions as encountered during its operational life cycle

NOTE Environmental tests cover natural and induced environments.

3.2.11 full functional test (FFT)

comprehensive test that demonstrates the integrity of all functions of the item under test, in all operational modes, including back-up modes and all foreseen transitions

NOTE 1 The main objectives of this test is to demonstrate absence of design manufacturing and integration error.

NOTE 2 FFT exists at the different level of decomposition of a space segment element. For satellite they also called system functional test (SFT) or integrated system test (IST).

3.2.12 maximum expected acceleration

acceleration value determined from the combined effects of the steady state acceleration and the transient response of the item as it will experience during its life time

NOTE 1 This term is equivalent to limit load (as defined in E-ST-32).

NOTE 2 Examples of events during life time are transportation, handling, engine ignition, engine burnout, and stage separation.

3.2.13 maximum expected acoustic spectrum

maximum value of the time average root-mean-square (r.m.s.) sound pressure level (SPL) in each frequency band occurring inside the payload fairing, orbiter, or cargo bay, which occurs during flight events

NOTE 1 E.g. lift-off, powered flight or re-entry.

NOTE 2 The maximum expected acoustic environment test spectrum is specified in octave or 1/3 octave bands over a frequency range of 31,5 Hz to 10 kHz. The duration of the maximum environment is the total period when the overall amplitude is within 6 dB of the maximum overall amplitude.

3.2.14 maximum expected shock

worst cases of the collection of the shock at their mounting interface due to every possible cause

NOTE 1 For example: causes of shocks are stage, shroud or satellite separation pyro elements, non-explosive actuators, mechanisms with energy release, appendage latching, and fuel valves.