
**Space systems — Spacecraft interface
requirements document for launch
vehicle services**

*Systèmes spatiaux — Document d'exigences d'interface du véhicule
spatial vis-à-vis du service de lancement*

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Foreword

ISO (the International Organization for Standardization) is a worldwide federation of national standards bodies (ISO member bodies). The work of preparing International Standards is normally carried out through ISO technical committees. Each member body interested in a subject for which a technical committee has been established has the right to be represented on that committee. International organizations, governmental and non-governmental, in liaison with ISO, also take part in the work. ISO collaborates closely with the International Electrotechnical Commission (IEC) on all matters of electrotechnical standardization.

The procedures used to develop this document and those intended for its further maintenance are described in the ISO/IEC Directives, Part 1. In particular, the different approval criteria needed for the different types of ISO documents should be noted. This document was drafted in accordance with the editorial rules of the ISO/IEC Directives, Part 2 (see www.iso.org/directives).

Attention is drawn to the possibility that some of the elements of this document may be the subject of patent rights. ISO shall not be held responsible for identifying any or all such patent rights. Details of any patent rights identified during the development of the document will be in the Introduction and/or on the ISO list of patent declarations received (see www.iso.org/patents).

Any trade name used in this document is information given for the convenience of users and does not constitute an endorsement.

For an explanation of the voluntary nature of standards, the meaning of ISO specific terms and expressions related to conformity assessment, as well as information about ISO's adherence to the World Trade Organization (WTO) principles in the Technical Barriers to Trade (TBT), see www.iso.org/iso/foreword.html.

This document was prepared by Technical Committee ISO/TC 20, *Aircraft and space vehicles*, Subcommittee SC 14, *Space systems and operations*.

This second edition cancels and replaces the first edition (ISO 17401:2004), which has been technically revised.

The main changes are as follows:

- upgrade orbit definition parameters;
- upgrade SC environment description;
- upgrade SC description (interface, thermal aspects, EMC).

Any feedback or questions on this document should be directed to the user's national standards body. A complete listing of these bodies can be found at www.iso.org/members.html.

Introduction

This document gives guidelines for writing an interface requirements document (IRD) for launch vehicle (LV) services. The application of this document is intended to facilitate the technical exchanges between spacecraft (SC) and launch vehicle agencies. By reducing the amount of work necessary for requesting launch services, this document will minimize spacecraft contractor's and spacecraft manufacturer's costs.

In some cases, drawings are explicitly requested in order to provide comprehensive information. Explicit international system units are specified for all items. The corresponding scale may be adjusted if not appropriate.

SC organizations may include additional topics if required. Some sections of the IRD may refer to specificities that are not applicable to the launch services of interest, in which case they should be ignored.

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Space systems — Spacecraft interface requirements document for launch vehicle services

1 Scope

This document provides spacecraft (SC) organizations with the general format for presenting the interface requirement document (IRD) for launch vehicle services. The IRD provides a list of the major technical requirements spacecraft agencies provide to launch vehicle (LV) agencies when submitting an application for launch services.

The IRD addresses the definition of the SC mission, the mechanical and electrical interfaces, the overall environment requirements (mechanical, thermal, cleanliness, radio-electrical), the SC development and test programme and, finally, launch range facilities and support requirements.

This document is applicable to all existing commercial LV and related launch facilities so as to permit SC contractors to prepare a single interface requirement document for a given SC mission, independently of the LV contractor to be selected.

The IRD, as defined in this document, includes the basic SC input data needed by LV agencies to prepare the interface control document defined in ISO 15863.

2 Normative references

There are no normative references in this document.

3 Terms, definitions and abbreviated terms

3.1 Terms and definitions

For the purposes of this document, the following terms and definitions apply.

ISO and IEC maintain terminology databases for use in standardization at the following addresses:

- ISO Online browsing platform: available at <https://www.iso.org/obp>
- IEC Electropedia: available at <https://www.electropedia.org/>

3.1.1

usable volume

volume available to the payload within the LV fairing or carrying structure that the static envelope of the SC may not exceed in order to ensure that there is no physical contact between the SC and the LV in a dynamic environment

3.1.2

spacecraft adapter

SC adapter

structure that mates the SC to the LV and includes the separation system for SC/LV separation

Note 1 to entry: The SC adapter is a part of the LV and does not separate with the SC.

3.2 Abbreviated terms

EIRP equivalent isotropic radiated power

EMC	electromagnetic compatibility
IRD	interface requirement document
LV	launch vehicle
RF	radio frequency
SC	spacecraft
Y/N	yes/no response

4 Spacecraft mission description

4.1 Mission description

The IRD shall describe the following items:

- a) purpose of the mission;
- b) orbital characteristics;
- c) in orbit view of SC, with drawing

4.2 SC description (optional)

The IRD may describe the following items:

- a) SC platform
- b) SC payload

5 Mechanical interfaces

5.1 Mechanical configuration

The IRD shall provide the following information:

- a) SC mechanical drawing (launch configuration);
- b) SC coordinate system, with drawing;
- c) maximum height above interface plane, in metres;
- d) SC overall dimensions (maximum diameter, maximum height, ...), in metres;
- e) SC/LV interface diameter, in metres. SC/LV interface geometrical definition.
- f) available volumes of SC, in cubic meters
- g) SC mass center position – relative to the reference frame which is centered in the adapter plane;
- h) mean-square deviation of SC mass center position along SC axis

5.2 SC fundamental frequencies

The IRD shall provide the fundamental natural frequencies of SC (specify boundary conditions):

- a) axial frequencies, in hertz;

- b) lateral frequencies, in hertz.

5.3 Usable volume

The IRD shall provide the following information:

- a) static envelope, with drawing;
- b) SC protrusions below interface plane, with dimensioned drawings;
- c) SC volumetric displacement, in cubic metres;
- d) SC free air volume, in cubic metres;
- e) special clearance requirements, with drawing.

5.4 Spacecraft (or SC adapter) mechanical interface

The IRD shall provide the following information:

- a) mechanical interface, with drawing;
- b) diameter, in metres;
- c) attachments at SC interface;
- d) material;
- e) Young's modulus, in newtons per square metre;
- f) coating:
 - 1) surfaces in contact,
 - 2) other surfaces;
- g) roughness, in micrometers of surfaces in contact;
- h) flatness/perpendicularity of surfaces in contact;
- i) stiffness (for clampband mating system):
 - 1) applicable length (height), in meters,
 - 2) section area, in square millimeters,
 - 3) inertia (with respect to centre of gravity of section), in millimeters to the power of 4 (mm⁴);
- j) stiffness (except for clampband mating systems):
 - 1) radial direction, in newtons per metre,
 - 2) tangent direction, in newtons per metre.

NOTE This section of the IRD applies to the lower adapter interface ring for a SC provided adapter.

5.5 Connectors and microswitches (SC side of the interface)

The IRD shall provide the following information:

- a) manufacturer and part number;
- b) quantity;

- c) location and mechanical interface, with drawing:
 - 1) angular position, in degrees,
 - 2) radial position, in metres,
 - 3) height from separation plane, in metres;
- d) push-on and push-off loads, in newtons;
- e) energy released, in joules;
- f) keying index.

5.6 Purges and fluid connection interface

The IRD shall provide the following information:

- a) definition;
- b) location and mechanical interface, with drawing:
 - 1) angular position, in degrees,
 - 2) radial position, in metres,
 - 3) height from separation plane, in metres.

5.7 Encapsulated spacecraft access

For each access requirement to payload, the IRD shall specify the following items:

- a) location in SC coordinates, in metres, with drawing:
 - 1) X_S ,
 - 2) Y_S ,
 - 3) Z_S ;
- b) minimum size of access door, length and height in metres;
- c) purpose.

NOTE The symbols X_S , Y_S , Z_S are the generic coordinates of any point in SC axes (or in axes parallel to the SC axes).

6 Electrical interfaces

6.1 Umbilical wiring diagram

Drawings for SC to LV and SC to ground facilities wiring shall be provided.

6.2 Umbilical connectors

The IRD shall provide the following information:

- a) number of connectors required;
- b) LV supplied (Y/N, to specify);
- c) manufacturer;

- d) connector type, name, designation and part number;
- e) number of pins needed for user;
- f) polarizing key type and orientation;
- g) insert key location, with drawing;
- h) location, with drawing;
- i) backshell shielding requirement;
- j) harness shielding requirement.

6.3 Umbilical wiring links (for each connector pin)

The IRD shall provide the following information:

- a) pin number;
- b) function(s);
- c) wire type;
- d) twisting and shielding characteristics;
- e) maximum voltage, in volts;
- f) maximum voltage at lift-off, in volts;
- g) maximum current, in amperes;
- h) maximum current at lift-off, in amperes;
- i) maximum one way resistance, in ohms;
- j) maximum voltage drop, in volts;
- k) line start point;
- l) line end point;
- m) maximum voltage at separation (if applicable), in volts;
- n) maximum current at separation (if applicable), in amperes;
- o) signal type;
- p) signal frequency, in hertz.

6.4 Electrical commands dedicated to spacecraft

6.4.1 Pyrotechnic commands

The IRD shall provide the following information:

- a) number of commands required;
- b) electrical circuit drawing, with drawing;
- c) command identification;
- d) number of initiators per command;

- e) time of command initiation;
- f) minimum time interval between commands, in seconds;
- g) pulse width, in seconds;
- h) voltage, in volts;
- i) minimum all fire current, in amperes;
- j) maximum no fire current, in amperes;
- k) output isolation, in ohms;
- l) wire gage;
- m) wire type;
- n) wire length from LV-SC interface to pyrotechnic devices, in metres;
- o) circuit connectors to pyrotechnic devices;
- p) initiator characteristics.

6.4.2 Dry loop commands

The IRD shall provide the following information:

- a) number of commands required;
- b) command identification;
- c) number of redundant commands;
- d) time of command initiation (on ground or in flight);
- e) resistance (ON/OFF configurations), in ohms;
- f) maximum, minimum and nominal output voltage, in volts;
- g) maximum current, in amperes;
- h) on-board circuit isolation, in ohms;
- i) grounding requirements;
- j) SC circuit configuration, with drawing.

6.4.3 Electrical commands

The IRD shall provide the following information:

- a) number of commands required;
- b) command identification
- c) number of redundant commands;
- d) time of command initiation (on ground or in flight);
- e) minimum time interval between commands, in seconds;
- f) maximum, minimum and nominal output voltage, in volts;
- g) maximum current, in amperes;